PERFORMANCE ANALYSIS OF MILITARY FLYING WING UAV WITH PULSE JET ENGINE

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Abstract: The current technological level of data acquisition systems, used on military and civilian unmanned aircraft, allow for diverse kinetic sampling conditions covering large ranges of altitudes and temperature. Thus, there is a need to collect the necessary atmospheric data, in terms of constructive simplicity and reduction of total costs (manufacturing and operation), with the help of unmanned aircraft equipped with PJE (pulse jet engine) and PDE (pulse detonation engine) propulsion systems. The article presents pre-design aspects of a flying wing UAV concept equipped with a PJE engine, which can be mounted on supersonic aircraft.

Keywords: flying wing, pulse jet engine, pulse detonation engine, XFLR5

Symbols and acronyms

AR	Aspect ratio	AoA	Angle of attack
CAD	Computer Aided Design	CAM	Computer Aided Manufacturing
CFD	Computational fluid dynamics	ISR	Intelligence, surveillance, and reconnaissance
LLT	Lifting line theory	MAC	Main aerodynamic chord
PDE	Pulse detonation engine	PJE	Pulse jet engine
VLM	Vortex Lattice Method		
c_0	Root chord	C_e	Tip chord
<i>c</i> _{<i>l</i>} ,	Lift coefficient	C_m	Pitch coefficient
c_d	Drag coefficient		

1. INTRODUCTION

PDEs are propulsion systems, in experimental phases, that use supersonic detonation waves as a combustion mechanism, having a reduced mechanical complexity and high efficiency compared to traditional aerofoil engines, PDE being considered an extension of PJE [5].

Due to the thermodynamic conditions of combustion, it can be an eligible candidate for equipping UAVs for subsonic and supersonic kinetic conditions, under the conditions of a UAV concept optimized for atmospheric data acquisition missions at high altitudes.

PDE can be an eligible technical solution for the propulsion of UAVs with fixed wing in classic concept or tailless (flying wing) in airborne launch conditions. Starting from the Fi 103 (V1) missile, historical and evolutionary milestones have confirmed the usefulness of (military) UAVs, such as: AQM-34 Firebee (Fig. 1.1), Lockheed D-21 (Fig. 1.2), BQM-74), equipped with no rotor engine-PJE engines.

The PDE concept offers a number of advantages, the most relevant being: technological and design simplicity, record theoretical power per volume unit, they are ecological, low impact on the engine and aircraft structure.

However, the concept of PDEs does not offer commercial applications with technological limitations still to be solved, such as: ignition of the detonation wave in a controllable and reliable way, geometric optimization of the detonation tube, cycle repetition frequency, optimization of manufacturing materials, [5]. Along with technological progress, the PDE concept is analysed multidisciplinary by the scientific community, a concept subject to concerns in the field of experimental thermodynamics and CFD analyses, [4].



FIG. 1.1 AQM-34 Firebee, [1]

FIG. 1.2 Lockheed D 21, [2]

According to specialist information [8, 9] a PDE engine was used on modified Scaled Composites-EZ (Fig. 1.3) thus demonstrating the efficiency of using such a propulsion system in terms of fuel consumption versus traction characteristics and of the level of atmospheric pollution. PJE engines are used in UAS and general and sport aviation, [10, 11].



FIG 1.3 Rutan Long-EZ Borealis cu PDE, [8, 9]

FIG 1.4 Glider PJE, [10]

2.AERODYNAMIC PERFORMANCE EVALUATION TOOLS AND METHODS

2.1. XFLR5 software

XFLR5 provides a series of useful modules for geometric parameterization and aerodynamic performance evaluation (2D and 3D) for aircraft without the influence and interference generated by propulsion systems (propellers). For XFLR5 6.55, the modules are as follows: direct foil design (geometric parameterization of aerofoils), Xfoil inverse design (generation of profiles from selected/edited aerofoils), Xdirect analysis (analysis of aerofoil performances), wing and plane design (geometric parameterization) and 3D aerodynamic analysis single geometries/complex geometries).

For 3D aerodynamic analysis the software tool can use a number of methods such as: LLT (lifting line theory), VLM (vortex lattice method) and 3D panels. The results

of the analyses are graphic, numerical with possibilities of poor processing and use with other software tools (e.g. CAD/CAM software, Excel), [7, 13].

2.2. Aerodynamic analysis methods

a. The VLM method

VLM (Vortex Latice Method) is an extension of the LLT (lifting line theory) method, based on the theory of ideal flow (potential flow), used both in the pre-design stages of aircraft (for a quick estimation of global aerodynamic characteristics) and in the field aerospace university and provides a reasonable level of accuracy. Through the initial estimates of the pressure distribution on the wing, the structural design phases of the fixed (wing, wings) and mobile (flaps, ailerons, thruster) bearing surfaces can be initiated, see Fig. 2.1.

VLM models fixed aerofoil surfaces (wing, tailplanes) as a thin web of discrete vortices for the calculation of lift and induced drag without the influence of wing thickness, turbulence, boundary layer and air viscosity. However, the method offers a global approach to the distribution of the pressure coefficient that determines the quantification of the relevant aerodynamic coefficients (and derived quantities) for possible evaluations of the stability/manoeuvrability qualities for the conceptual phase of the aircraft, [17, 18].

The method is described clearly and in detail in a series of specialized references in the field of aerodynamics, the most relevant being: Katz-Plotkin [19], Anderson [20], Bertin-Simth [21] or Drela [22].



FIG 2.1 VLM method a.[18], b.[23]

b. The 3D panel method

This method analyses the distribution of the pressure coefficient (intrados/extrados) of a bearing surface also considering the thickness of the wing (3D geometry) compared to the VLM which was based only on the curvature line of the wing (2D geometry), see Fig. 2.2.

3D panel models the perturbation existing on the wing surface, initially being an analysis method with singularities (source distributions and vortices) later being a variant of boundary element methods. Currently this method can be used for a solution of the flow potential equation on complex geometries for subsonic or supersonic speeds, the method eliminating the need for a volumetric grid as in the case of finite element methods, [18].



FIG 2.2 3D panel method, geometry approximation by panel elements [19]

3.CONCEPT FLYING WING UAV-PJE

The UAV-PJE concept is realized with the software tool XFLR5 [6, 7] based on a flying wing geometry (see Fig. 2.1) that can be used in ISR missions, powered by a miniPJE engine, having the characteristics estimated in Table 2.1.



FIG. 2.1 UAV-PJE/PDE initial proposal concept

Characteristics	Value	Characteristics	Value				
Span / Length	2 / 1,5 m	MAC	0,35 m				
Dihedral / Swept	5° / 26,79°	Wing load	8,1 kg/m ²				
Surface	$0,574 \text{ m}^2$	Total weight/ Payload	4,65 kg / 1,25 kg				
Aspect ratio / taper ratio	6,97 / 0,171	Propulsion	PJE / pulse-jet				

Table 2.1 Initial estimated characteristics for FW UAV-PJE

The estimation of the mass and balance variant is highlighted graphically in Fig. 2.2 and numerically in Table 2.2 and annex 3.

		Table 2.2 FW UAV-PJE weigh	t ana balance
Element	Value	Element	Value
Frame weight	2,2 kg	Fuel	0,5 kg
Engine weight	1,2 kg	Battery	0,25 kg
Electronic equipment weight	0,5	Total weight	4,65 kg

Table 2.2 FW UAV-PJE weight and balance



FIG. 2.2 UAV-PJE weight and balance for initial proposal concept

To make a 3D model, XFLR5 can export *.stl files used for 3D printing with the Cura 15 tool, see Fig. 2.3, [15]. The 3D model can be used to measure aerodynamic parameters in subsonic wind tunnels, [16].





4.UAV-PJE PERFORMANCES ANALYSIS

4.1. Simulation conditions and analysis cases

For optimal parametrisation from the point of view of the useful space of the wing, a high-thickness profile specific to the MH 92 flying wings was chosen for the fitting, and for the extreme section the MH 64 was selected, thus generating a mixed geometric torsion, both designed by Martin Hepperle, see Fig. 3.1, having the characteristics in Table 3.1, [12].



FIG. 3.1 Aerofoils [13]

Table 3.1 Aerofoil features

	MH 92	MH 64			
Max. thickness	15,50% la 27,5%	Max. thickness	8,6% la 26,9%		
Max. curvature	1,4% la 15%	Max. curvature	1,4% la 41,8%		

4.2. Performance analysis for 2D geometry (XFLR5)

The simulation steps aim at an estimation of the aerodynamic performance of the proposed geometric configuration (see Table 2.1). according to the analysis conditions recorded in Table 3.2.

		Table 3.1	2 2D analysis conditions
Analysis condition	Value	Analysis condition	Value
Chord	1m	Speed	15 m/s
Nr. Reynolds	1100000	Air density	1,225 kg/m ³
Iterations	100	Aer viscosity	1,5 x 10 ⁻⁵ m ² /s

The analysed aerofoils provide the variation of comparative values over the range of AoA analysed to observe the aerodynamic behaviour at a minimum theoretical speed (15 m/s).



FIG. 3.2 Airfoil polars, a.c₁ vs AoA, b.c_d vs AoA, [13]

Fig. 3.2 and 3.3 show both the polars of the relevant aerodynamic coefficients c_1 , c_d and c_m as a function of AoA, as well as the gliding rate c_1/c_d vs AoA at the analysed speed (Fig. 3.2c) or the mutual variations of c_1 vs c_d (Fig. 3.2.d).



FIG. 3.3 Aerofoil polars, c. c_1/c_d vs AoA, d. c_1 vs c_d [13]



The differences in the values of the aerodynamic coefficients for the two analysed aerofoils recommend their use in the span: MH 92 at the embedment and MH 64 at the tip of the wing. The pitching moment differences (Fig. 3.4.a) recommend an optimization of the overall performance through aerodynamic torsion (see also annex 1).

4.3. 3D geometry performance analysis

It is focused on a series of aerodynamic analyses for the single flying wing without the presence and influence of the fuselage in the form of aerodynamic interferences, having the atmospheric and kinetic analysis conditions from Table 3.3.

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		Table 3	3 3D analysis conditions
Analysis condition	Value	Analysis condition	Value
Span	2 m	Speed	15 m/s
Surface	$0,57 \text{ m}^2$	Boundary conditions	Dirichlet
AoA	$-15^{\circ} \div 15^{\circ}$	AoA precision	0,01
Method of analysis	VLM/3D panel	Panels VLM / 3D	494 / 1014

The analysed 3D geometry (see also Table 2.1) provides a series of aerodynamic parameter values, highlighted in Fig. 3.5 and 3.6 and Annex 2.



FIG. 3.5 UAV-PJE flying wing polars, a.Cl-AoA, b.Cd-AoA [13]



FIG. 3.6 UAV-PJE flying wing polars, c.C₁-C_d, d.C_m-AoA [13]

According to Fig. 3.5 and 3.6 differences in the values of the aerodynamic coefficients are observed with the increase of AoA due to the consideration of the friction effect in the case of the 3D panel method.



FIG. 3.7 Drag force (AoA=9°), a.VLM method, b. 3D panel method

In Fig. 3.7, we have the comparative values of the drag for a critical value $AoA=9^{\circ}$, and can be observed the trace of the drag generated by the air viscosity. In Fig. 3.8 the values of the pressure coefficient Cp extracted with the two analysis methods: VLM and 2D panel are observed, generated value differences generated by the consideration of air viscosity.



FIG. 3.8 Pressure coefficient distribution C_p (AoA=9°), a.VLM method, b. 3D panel method

5. PROPULSION SYSTEM (PJE) PERFORMANCE ESTIMATE

A mass-optimized PJE for the analysed UAV (maximum weight of 4.65 kg) has the dimensions estimated from Table 5.1, values obtained with Pulse Jet Engine Calculator 1.4 [24]. Figure 5.1. shows a simple PJE geometry with the main dimensions calculated.



FIG. 5.1 Pulse jet engine a.geometry a.[24], b. CAD aspects,[25]

Features	Value	Features	Value
Lenght C	68 cm	Valve head lenght	10.21 cm
Lenght combustor D	13,62 cm	Exhaust cone E	13,62
Exhaust pipe F	40,86	Detonation freguency	189-203 Hz
Exhaust B	4,75 cm	Thrust	5kg/cm

Table 5.1 PJE structural features and performances estimated

CONCLUSIONS

Currently, PJE-powered unmanned aircraft are a mature technology that can offer advantages to users in areas of interest, technology that is also refined with the help of analysis software tools.

The article summarized a pre-design stage of a PJE powered flying wing UAV using freeware tools (XFLR5, PJE calculator).

The current study can be substantiated with the help of CFD tools that can provide high degrees of confidence both in problematic kinetic cases (transonic speeds) and in manoeuvrings flight situations (high angles of attack, high bank angles). In the future, it is considered that the CFD instrumentation will be applied to virtual and real geometric models for a comparative approach to the results and the choice of optimized technical solutions.

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Anexx 1. 2D data export

xflr5 v6.	55										
Calculat	ed polar f	For: MH64 @	0 Martin H	epperle							
1 1 Reyn	olds numbe	er fixed	Ма	ch numbe	r fixed						
xtrf =	1.000 (to	op)	1.000 (bo	ttom)							
Mach =	0.044	Re =	1.100 e 6	Ncn:	it = 9	.000					
alpha	CL	CD	СОр	Cm	Top Xtr	Bot Xtr	Cpmin	Chinge	ХСр	-	
-7.000	-0.6218	0.03382	0.03139	-0.0212	0.9526	0.0090	-4.1838	0.0000	0.0000	0.0000	0.2118
-6.000	-0.5487	0.02674	0.02354	-0.0187	0.8810	0.0118	-3.3770	0.0000	0.0000	0.0000	0.2123
-5.000	-0.4446	0.02540	0.02174	-0.0170	0.8384	0.0182	-2.4238	0.0000	0.0000	0.0000	0.2087
-3.000	-0.2430	0.00847	0.00268	-0.0113	0.7659	0.0118	-1.9976	0.0000	0.0000	0.0000	0.2008
-1.000	-0.0250	0.00539	0.000113	-0.0098	0.6967	0.4759	-0.4855	0.0000	0.0000	0.0000	-0.1481
0.000	0.0844	0.00481	0.00072	-0.0091	0.6512	0.6780	-0.4154	0.0000	0.0000	0.0000	0.3564
1.000	0.2092	0.00429	0.00077	-0.0112	0.5831	0.9778	-0.5479	0.0000	0.0000	0.0000	0.3017
2.000	0.3369	0.00472	0.00084	-0.0147	0.4934	1.0000	-0.7241	0.0000	0.0000	0.0000	0.2914
3.000	0.4450	0.00534	0.00111	-0.0140	0.3959	1.0000	-0.9345	0.0000	0.0000	0.0000	0.2787
4.000	0.5529	0.00609	0.00152	-0.0135	0.3038	1.0000	-1.2262	0.0000	0.0000	0.0000	0.2710
5.000	0.6600	0.00711	0.00213	-0.0130	0.2057	1.0000	-2.0487	0.0000	0.0000	0.0000	0.2657
6.000	0.7660	0.00835	0.00301	-0.0126	0.1190	1.0000	-3.1175	0.0000	0.0000	0.0000	0.2615
8.000	1 0400	0.01457	0.00941	-0.0092	0.0083	1.0000	-5.0423	0.0000	0.0000	0.0000	0.2525
10.000	1.1297	0.02195	0.01763	-0.0016	0.0009	1.0000	-8.2534	0.0000	0.0000	0.0000	0.2445
11.000	1.2202	0.02312	0.01887	-0.0034	0.0032	1.0000	-10.4633	0.0000	0.0000	0.0000	0.2417
12.000	1.2263	0.03184	0.02838	0.0048	0.0022	1.0000	-10.9410	0.0000	0.0000	0.0000	0.2336
13.000	1.1720	0.04812	0.04547	0.0039	0.0022	1.0000	-10.7108	0.0000	0.0000	0.0000	0.2326
14.000	1.0972	0.07533	0.07332	-0.0122	0.0022	1.0000	-10.0683	0.0000	0.0000	0.0000	0.2454
15.000	1.0107	0.11272	0.11115	-0.0350	0.0024	1.0000	-8.8762	0.0000	0.0000	0.0000	0.2680
Calcu 1 1 R	lated polar eynolds num	r for: MH92 nber fixed	© Martin M	Hepperle ach numbe	er fixed						
xtrf Mach	= 1.000 (= 0.044	(top) Re =	1.000 (b 1.100 e	ottom) 6 Ncr	it = 9	.000					
alph	a CL	CD	CDp	Cm	Top Xtr	Bot Xtr	Cpmin	Chinge	ХСр		
-14.0	00 -0.5713	0.15597	0.15438	0.0374	1.0000	0.0181	-1.8024	0.0000	0.0000	0.0000	0.2984
-13.0	00 -0.5590	0.13769	0.13613	0.0282	1.0000	0.0181	-1.9795	0.0000	0.0000	0.0000	0.2840
-12.0	00 -0.5705 20 -0.0057	0.11241	0.11088	0.0156	1.0000	0.0180	-2.3847	0.0000	0.0000	0.0000	0.2017
-11.0	00 -0.9933	0.01788	0.01236	0.0111	0.8228	0.0058	-5.4607	0.0000	0.0000	0.0000	0.2516
-8.0	00 -0.8524	0.01328	0.00706	0.0198	0.7689	0.0120	-4.9014	0.0000	0.0000	0.0000	0.2636
-7.0	00 -0.7477	0.01264	0.00641	0.0213	0.7209	0.0158	-3.8884	0.0000	0.0000	0.0000	0.2697
-4.0	00 -0.4261	L 0.01055	0.00391	0.0247	0.5863	0.0197	-1.8045	0.0000	0.0000	0.0000	0.3011
-2.0	00 -0.2339	0.00663	0.00161	0.0304	0.4939	0.5185	-1.0432	0.0000	0.0000	0.0000	0.3741
-1.0	00 -0.1207 00 -0.01207	0.00604	0.00145	0.0317	0.4580	0.0/9/	-0.01/1	0.0000	0.0000	0.0000	0.4955
1.0	0.1012	3 0.00620	0.00140	0.0318	0.3821	0.7959	-0.8780	0.0000	0.0000	0.0000 -	0.0723
2.0	0.2176	0.00653	0.00178	0.0315	0.3422	0.8242	-1.0323	0.0000	0.0000	0.0000	0.0964
3.0	0.3314	1 0.00698	0.00207	0.0312	0.3071	0.8461	-1.1945	0.0000	0.0000	0.0000	0.1473
4.0	0.4454	0.00726	0.00242	0.0310	0.2901	0.8713	-1.3616	0.0000	0.0000	0.0000	0.1714
5.0	0.5575	5 0.00789	0.00287	0.0307	0.2562	0.8851	-1.5517	0.0000	0.0000	0.0000	0.1852
6.0	00 0.6696	5 0.00822	0.00326	0.0306	0.2444	0.9051	-1.7558	0.0000	0.0000	0.0000	0.1938
7.0	00 0.770 00 0.8810	0.00890	0.00597	0.0311	0.2135	0.9254	-2.0012	0.0000	0.0000	0.0000	0.1985
9.0	0.9826	0.01031	0.00531	0.0333	0.1772	0.9609	-2.5862	0.0000	0.0000	0.0000	0.2023
10.0	00 1.0914	0.01110	0.00621	0.0324	0.1650	0.9789	-2.9549	0.0000	0.0000	0.0000	0.2051
11.0	00 1.2154	0.01256	0.00753	0.0274	0.1323	0.9910	-3.3795	0.0000	0.0000	0.0000	0.2109
12.0	00 1.3158	3 0.01452	0.00931	0.0257	0.1010	1.0000	-3.8266	0.0000	0.0000	0.0000	0.2122
13.0	00 1.3636	5 0.01672	0.01147	0.0330	0.0810	1.0000	-4.2314	0.0000	0.0000	0.0000	0.2058
14.0	00 1.3933	6 0.02100	0.01579	0.0380	0.0616	1.0000	-4.7089	0.0000	0.0000	0.0000	0.2007
12.0	00 1.4180	0.02/0/	0.02198	0.0397	0.0526	1.0000	-5.2094	0.0000	0.0000	0.0000	0.19//

Anexx 2. 3D data export

lane name : olar name : reestream :	speed :	fw-pde engin T1-15.0 m/s- 15.000 m/s	e VLM1									
alpha	Beta	CL	CDi	CDv	CD	СҮ	c1	Ст	Cn	Cni	QInf	X
-15.000	0.000	-1.063660	0.051984	0.000000	0.051984	-0.000000	-0.000000	1.205421	-0.000000	-0.000000	15.0000	0.4
-14.000	0.000	-0.994501	0.045562	0.000000	0.045562	-0.000000	0.000000	1.130646	0.000000	0.000000	15.0000	0.4
-12.000	0.000	-0.854296	0.033807	0.000000	0.039490	-0.000000	0.000000	0.976965	-0.000000	-0.000000	15.0000	0.4
-11.000	0.000	-0.783334	0.028511	0.000000	0.028511	-0.000000	-0.000000	0.898246	0.000000	0.000000	15.0000	0.4
-10.000	0.000	-0.711856	0.023627	0.000000	0.023627	-0.000000	-0.000000	0.818402	-0.000000	-0.000000	15.0000	0.4
-9.000	0.000	-0.639907	0.019169	0.000000	0.019169	-0.000000	0.000000	0.737531	-0.000000	-0.000000	15.0000	0.4
-8,000	0.000	-0.567531	0.015152	0.000000	0.015152	-0.000000	0.000000	0.655730	0.000000	0.000000	15.0000	0.4
-7.000	0.000	-0.494773	0.011587	0.000000	0.011587	-0.000000	0.000000	0.573100	-0.000000	-0.000000	15.0000	0.4
-5.000	0.000	-0.348301	0.005862	0.000000	0.005862	-0.000000	0.000000	0.405756	-0.000000	-0.000000	15.0000	0.4
-4.000	0.000	-0.274682	0.003719	0.000000	0.003719	-0.000000	0.000000	0.321245	-0.000000	-0.000000	15.0000	0.4
-3.000	0.000	-0.200870	0.002066	0.000000	0.002066	-0.000000	0.000000	0.236312	-0.000000	-0.000000	15.0000	0.4
-2.000	0.000	-0.126914	0.000909	0.000000	0.000909	-0.000000	0.000000	0.151061	-0.000000	-0.000000	15.0000	0.4
-1.000	0.000	-0.052863	0.000251	0.000000	0.000251	0.000000	0.000000	0.065596	-0.000000	-0.000000	15.0000	0.4
0.000	0.000	0.021234	0.000095	0.000000	0.000095	0.000000	0.000000	-0.019980	0.000000	0.000000	15.0000	0.3
1.000	0.000	0.095329	0.000442	0.000000	0.000442	0.000000	0.000000	-0.105563	0.000000	0.000000	15,0000	0.4
3,000	0.000	0.243316	0.001291	0.000000	0.001291	0.000000	0.000000	-0.191047	0.000000	0.000000	15.0000	0.4
4.000	0.000	0.317111	0.004486	0.000000	0.004486	-0.000000	0.000000	-0.361304	0.000000	0.000000	15.0000	0.4
5.000	0.000	0.390708	0.006823	0.000000	0.006823	-0.000000	0.000000	-0.445870	0.000000	0.000000	15.0000	0.4
6.000	0.000	0.464060	0.009644	0.000000	0.009644	-0.000000	0.000000	-0.529924	0.000000	0.000000	15.0000	0.4
7.000	0.000	0.537119	0.012943	0.000000	0.012943	-0.000000	0.000000	-0.613362	0.000000	0.000000	15.0000	0.4
8.000	0.000	0.609838	0.016708	0.000000	0.016708	-0.000000	0.000000	-0.696083	0.000000	0.000000	15.0000	0.4
9.000	0.000	0.682170	0.020929	0.000000	0.020929	-0.000000	0.000000	-0.777987	0.000000	0.000000	15.0000	0.4
	0.000	0.754070	0.025593	0.000000	0.020093	-0.000000	0.000000	-0.858974	0.000000	0.000000	15.0000	0.4
11 000	0 000	V . U.L	0.030000	0.000000	0.026107	-0.000000	0.000000	-1.017801	0.000000	0.000000	15.0000	0.4
11.000	0.000	0.896392	0.036197	0.000000	0.03019/							
11.000 12.000 13.000	0.000 0.000 0.000	0.896392	0.036197	0.000000	0.042104	-0.000000	0.000000	-1.095449	0.00000	0.000000	15.0000	0.4
11.000 12.000 13.000 14.000	0.000 0.000 0.000 0.000	0.896392 0.966727 1.036454	0.036197 0.042104 0.048391	0.000000 0.000000 0.000000	0.042104 0.048391	-0.000000	0.000000	-1.095449	0.000000	0.000000	15.0000 15.0000	0.4
11.000 12.000 13.000 14.000 15.000 xflr5 v6. Plane nam Polar nam	0.000 0.000 0.000 0.000 0.000 55 55 te : te : te : te :	0.896392 0.966727 1.036454 1.105531 fw-pde eng: T1-15.0 m/s	0.036197 0.042104 0.048391 0.055040 ine s-Panel	0.000000 0.000000 0.000000 0.000000	0.042104 0.048391 0.055040	-0.000000 -0.000000 -0.000000	0.000000 0.000000 0.000000	-1.095449 -1.171792 -1.246739	0.000000 0.000000 0.000000	0.000000 0.000000 0.000000	15.0000 15.0000 15.0000	0.4 0.4
11.000 11.000 12.000 13.000 14.000 15.000 xflr5 v6. Plane nam Polar nam Freestrea	0.000 0.000 0.000 0.000 0.000 55 55 se : se : se : se : speed	0.896392 0.966727 1.036454 1.105531 fw-pde eng T1-15.0 m/s	0.036197 0.042104 0.048391 0.055040	0.000000 0.000000 0.000000 0.000000 0.000000	0.042104 0.042391 0.055040	-0.00000 -0.00000 -0.000000 -0.000000	0.00000 0.000000 0.000000	-1.095449 -1.171792 -1.246739	0,00000 0,00000 0,000000 0,000000	0.000000	15.0000 15.0000 15.0000	0.4 0.4 0.4
11.000 11.000 12.000 13.000 14.000 15.000 xflr5 v6. Plane nam Polar nam Freestrea	0.000 0.000 0.000 0.000 0.000 55 se : se : mm speed Bet:	0.896392 0.966727 1.036454 1.105531 fw-pde eng: T1-15.0 m/s 15.000 m/s	0.036197 0.042104 0.048391 0.055040 ine s-Panel CDi	0,000000 0,000000 0,000000 0,000000 0,000000	0.042104 0.042104 0.048391 0.055040	-0.000000 -0.000000 -0.000000 -0.000000	6.00000 6.000000 6.000000 6.000000	-1.095449 -1.171792 -1.246739 Cm 1.271620	0.00000 0.00000 0.000000 0.000000	0.000000 0.000000 0.000000 0.000000	QInf	0.4 0.4 0.4
11.000 12.000 13.000 14.000 15.000 xflr5 v6. Plane nam Polar nam Freestrea alpha -15.000 -14.000	0.000 0.000 0.000 0.000 0.000 55 se : re : mm speed Bet: 0.000 0.000	0.896392 0.966727 1.036454 1.105531 fw-pde eng: T1-15.0 m/s 15.000 m/s CL 10 -1.086837 0 -1015381	0.036197 0.042104 0.048391 0.055040 ine s-Panel CDi 0.054767 0.047916	0.000000 0.000000 0.000000 0.000000 0.000000	CD 0.150418 0.042104 0.042104 0.042104 0.055040	-0.000000 -0.000000 -0.000000 -0.000000 0.000000 0.000000	6.000000 6.000000 6.000000 6.000000 9.000000 9.000000	-1.095449 -1.171792 -1.246739 (m 1.271639 1.193181	0.00000 0.000000 0.000000 0.000000 -0.000000	0.000000 0.000000 0.000000 0.000000 0.000000	QInf 15.0000 15.0000 15.0000	0.4 0.4 0.4 0.4 0.4 2 0.4 2 8 0.4 2 8
10.000 11.000 12.000 13.000 14.000 15.000 xflr5 v6. Plane nam Polar nam Freestrea alpha -15.000 -14.000 -13.000	0.000 0.000 0.000 0.000 0.000 55 55 he : te : te : Bet: Bet: 0.00 0.000 0.000	0.896392 0.966727 1.036454 1.105531 T1-15.0 m/f 15.000 m/s 4 CL 10 -1.086837 90 -1.015881 0 -0.943261	0.036197 0.042104 0.048391 0.055040 ine s-Panel CDi 0.054767 0.047916 0.047916	0.000000 0.000000 0.000000 0.000000 0.000000	CD 0.150418 0.042104 0.042104 0.042104 0.055040 0.055040 0.150418 0.143567 0.126243	-0.000000 -0.000000 -0.000000 -0.000000 0.000000 0.000000 0.000000	0.000000 0.000000 0.000000 0.000000 0.000000	-1.095449 -1.171792 -1.246739 -1.246739 1.271639 1.193181 1.111920	0.00000 0.000000 0.000000 0.000000 -0.00000 -0.00000	0.000000 0.000000 0.000000 0.000000 0.000000	QInf 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000	0.4 0.4 0.4 0.4 0.4 273 0.4282 0.4291
10.000 11.000 12.000 13.000 15.000 xflr5 v6. Plane nam Polar nam Freestrea alpha -15.000 -14.000 -13.000 -12.000	0.000 0.000 0.000 0.000 0.000 55 55 Ne : : : : : : : : : : : : : : : : : : :	0.896392 0.966727 1.036654 1.105531 fw-pde eng T1-15.0 m/s 15.000 m/s L CL 0 -1.086837 0 -1.085831 0 -0.943261 0 -0.870520	0.036197 0.042104 0.048391 0.055640 ine s-Panel CDi 0.054767 0.047916 0.047916 0.047455 0.035403	0.000000 0.000000 0.000000 0.000000 0.000000	CD 0.159418 0.055040 0.159418 0.159418 0.159418 0.126243 0.110078	-0.000000 -0.000000 -0.000000 -0.000000 0.000000 0.000000 0.000000	0.000000 0.000000 0.000000 0.000000 0.000000	-1.095449 -1.171792 -1.246739 .1.246739 1.193181 1.111920 1.029252	0.000000 0.000000 0.000000 0.000000 -0.000000 -0.000000 -0.000000	0.000000 0.000000 0.000000 0.000000 0.000000	QInf 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000	0.4 0.4 0.4 0.4 273 0.4282 0.4291 0.4300
10.000 11.000 12.000 13.000 14.000 15.000 15.000 ranse Freestrea 15.000 -14.000 -14.000 -12.000 -12.000	0.000 0.000 0.000 0.000 0.000 0.000 55 55 He : He : He : He : He : He : He : He :	0.896392 0.966727 1.036454 1.105531 fw-pde eng: T1-15.0 m/s 15.000 m/s 6 CL 0 - 1.086837 N0 -1.015381 0 -0.943261 0 -0.870520 0 -0.773264	0.035197 0.042104 0.048391 0.055040 ine 5-Panel CD1 0.054767 0.047916 0.035403 0.035403	CDv e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.0000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.000000 e.0000000 e.00000000	CD 0.150419 0.042104 0.048391 0.055040 0.150418 0.143567 0.126243 0.110078 0.10078	-0.000000 -0.000000 -0.000000 -0.000000 0.000000 0.000000 0.000000 0.000000	0.000000 0.000000 0.000000 0.000000 0.000000	-1.095449 -1.171792 -1.246739 -1.246739 1.271639 1.193181 1.111920 1.02252 0.945320 0.945320	0.000000 0.000000 0.000000 0.000000 -0.000000 -0.000000 -0.000000 -0.000000	0.000000 0.000000 0.000000 0.000000 0.000000	QInf 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000	0.4 0.4 0.4 0.4 0.4 0.4 0.4 0.4 0.4 0.4
10.000 11.000 12.000 13.000 14.000 xflr5 v6. Plane nam Freestrea alpha -15.000 -14.000 -13.000 -12.000 -11.000 -10.000 -9.000	0.000 0.000 0.000 0.000 0.000 55 55 be : te : te : te : te : te : te : te : t	0.896392 0.966727 1.036454 1.105531 fw-pde eng: T1-15.0 m/s 15.000 m/s 4 CL 90 -1.086837 0 -1.015381 10 -0.943261 10 -0.879520 0 -0.773354 0 -0.773354 0 -0.773354	0.036197 0.042104 0.048391 0.055040 	0.000000 0.000000 0.000000 0.000000 0.000000	CD 0.159418 0.482391 0.055040 CD 0.159418 0.143567 0.126243 0.110978 0.20592 0.081148 0.405992 0.081148	-0.00000 -0.00000 -0.00000 -0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.00000 0.000000	0.000000 0.000000 0.000000 0.000000 0.000000	-1.095449 -1.171792 -1.246739 -1.246739 -1.246739 -1.271639 1.193181 1.111920 1.029252 0.945320 6.860237 0.724660	0.00000 0.000000 0.000000 0.000000 -0.00000 -0.00000 -0.00000 -0.00000 -0.00000	0.000000 0.000000 0.000000 0.000000 0.000000	QInf 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000 15.0000	0.4 0.4 0.4 0.4 0.4 0.4 0.4 0.4 0.4 0.4
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Plane nam Polar nam Freestrea alpha -15.000 -14.000 -12.000 -12.000 -10.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 -0.000 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#fw-pde engine1.m	ass - Notepad							•		
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<pre>#</pre>	ne unit and p ia breakdo 0010 m 0000 kg	arameter da wn.	ta.							
<pre>#- # Gravity and # Must be in t g = 9.81 rho = 1.225 #</pre>	density to the unit na tem's iner here must entation,	- be used as mes given a - tias about be exactly same origin	default bove (i.e item's ow the same location	values in tr . m,kg,s). n CG. one used in , same leng	rim setup (s n the .avl i th units)	saves runtime input file	typing).			
# # mass	×	V	7	Txx	TVV	T77	TXV	Tx7	Tv7	
1	403 -7	,15e-15	0	0.103	0.0201	0.124	0	0	- ,-	Main Wing
1.2	335	0	29.4	0.000232	0.0405	0.0402	0	0.00102	0	Body's inertia
0.25	350	-200	0	0.000	0.000	0.000 ! f	uel			
0.25	350	200	0	0.000	0.000	0.000 ! f	uel			
1.2	400	0	0	0.000	0.000	0.000 ! P	JE			
0.5	100	0	0	0.000	0.000	0.000 ! e	quipment			
0.25	150	0	0	0.000	0.000	0.000 ! b	aterry			
#		-								